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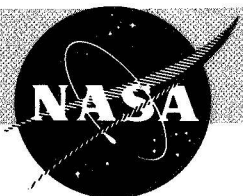


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FY 1970 SCIENTIFIC AND TECHNICAL REPORTS
ARTICLES, PAPERS, AND PRESENTATIONS

Management Services Office

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION



ABSTRACT

This document presents formal NASA technical reports, papers published in technical journals, and presentations by MSFC personnel in FY 70. It also includes papers of MSFC contractors.

After being announced in STAR or C STAR, all of the NASA series reports listed may be obtained from the Scientific and Technical Information Facility, P. O. Box 33, College Park, Maryland 20740.

The information in this report will be of value to the scientific and engineering community in determining what information has been published and what is available.

A&TS-MS-I-70-01

September 1970

FY 1970 SCIENTIFIC AND TECHNICAL REPORTS,
ARTICLES, PAPERS, AND PRESENTATIONS

Scientific and Technical Information Division
Management Services Office

FOREWORD

In accordance with the NASA Space Act of 1958 the MSFC has provided for the widest practicable and appropriate dissemination of information concerning its activities and the results thereof.

Since July 1, 1960, when the George C. Marshall Space Flight Center was organized, the reporting of scientific and engineering information has been considered a prime responsibility of the Center. Our credo has been that "research and development work is valuable, but only if its results can be communicated and made understandable to others."

GEORGE C. MARSHALL SPACE FLIGHT CENTER
Marshall Space Flight Center, Alabama

FY 1970 SCIENTIFIC AND TECHNICAL REPORTS
ARTICLES, PAPERS, AND PRESENTATIONS

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NASA TECHNICAL MEMORANDA

TM X-53859

July 8, 1969

Lunar Excursion Module RCS Engine Vacuum Chamber Contamination Study. Gary M. Arnett, Technical Coordinator. Space Sciences Laboratory. N70-13418.

The objective of this study was the definition of future contamination studies and procedures, and the investigation extended to the effects of the Reaction Control System (RCS) plume on optical flight experiments. This report is concerned with the effects of the RCS plume deposits on the test beds along with the characteristic changes that occur once these deposits are exposed to the laboratory atmosphere. Test beds consisting of various optical surfaces were exposed to an LM-RCS rocket engine plume in the vacuum chamber A. Analysis of the contaminated test beds included optical measurements ranging from the near-ultraviolet through the far-infrared region together with mass spectrometer identification of the deposits.

TM X-53865

October 31, 1969

Natural Environment Criteria for the NASA Space Station Program. Don K. Weidner, ed. Aero-Astrodynamics Laboratory.

This document provides natural environment criteria for the NASA Space Station Program with emphasis on the 1975-1985 time period. Information in the disciplinary areas of atmospheric and ionospheric properties, radiation, solar cycle predictions, geomagnetic field, astrodynamics constant, and meteoroids is given for the Earth, Moon, Mars, Venus, and interplanetary space. This document, originally published on August 1, 1969, has been revised to include the planetary and lunar environment criteria needed for preliminary Space Station studies relative to a manned Mars landing, a Venus flyby, and a lunar mission. It supersedes earlier versions of NASA TM X-53865.

TM X-53866

August 12, 1969

Z Transform and the Use of the Digital Differential Analyzer as a Peripheral Device to a General Purpose Computer. Donald Nalley. Computation Laboratory. N70-19896.

A feasibility study determined whether the Digital Differential Analyzer (DDA) can be used as a peripheral device; that is, a computer system programmed in FORTRAN IV or some other standard language in which certain manipulations could be accomplished on the DDA by calling a subroutine. In this study, consideration was given to Large Scale Integrator Circuits (LSI), because it was felt that suitable LSI would make the DDA a more feasible device.

It appears that the DDA as a standard peripheral device with automatic patching is indeed feasible. However, implementation with LSI is questionable presently, but this may become feasible soon.

TM X-53867

May 12, 1970

Probability Distributions for Thunderstorm Activity at Cape Kennedy, Florida. Lee W. Falls, William O. Williford, and Michael C. Carter. Aero-Astrodynamics Laboratory.

Several statistical distribution functions are investigated as prospective models to represent the variation of thunderstorm activity at Cape Kennedy, Florida. Statistical methods are presented using the latest and most comprehensive thunderstorm data available. The conclusion is reached that the negative binomial distribution and a modification of the negative binomial distribution are adequate statistical models to represent "thunderstorm events" and "thunderstorm hits," respectively, at Cape Kennedy, Florida.

TM X-53868

August 27, 1969

Studies in System Simulation. R. Saeks. NASA-ASEE Summer Faculty Fellowship Program. N69-41152.

The results of three mathematical studies of the feasibility of various methods of large scale digital system simulation are reported. Specific studies deal with the effects of applying numerical techniques directly to the components of a system, approximation techniques for reducing the order of subsystems, and data compression techniques for the computer storage of the large topological matrices encountered in system simulation.

NASA TECHNICAL MEMORANDA

TM X-53870

August 26, 1969

A Direct Measurement of the Most Probable Preferred Angular Velocity of Turbulent Structures by Optical Correlation of Laser Schlieren Signals. B. H. Funk. Aero-Astro dynamics Laboratory. N70-12209.

A method which potentially provides the means for separating the translational and rotational motion of turbulent structures is introduced. Simple two-dimensional models are used to relate the skewness of cross-correlograms computed from laser schlieren signals to the rotation of flow disturbances. The method, referred to herein as the method of "forced similarity," is discussed with respect to application to the turbulent free shear layer of an axisymmetric supersonic jet. Experimental results show that the shape of the cross-correlogram in the neighborhood of the "peak" is strongly influenced by rotational motion, and therefore, it becomes necessary to account for the effect in order to determine the correct statistical properties of the turbulence.

TM X-53871

September 5, 1969

Experiments in Hydrodynamic Grease Bearings. John L. Burch and Peter H. Broussard, Jr. Astrionics Laboratory. N70-11874.

The suitability of using hydrodynamic conical- and spherical-shaped spiral-grooved grease bearings as a replacement for gyro ball bearings has been investigated. Grease bearings have been built and tested for rotors which vary in diameter from a few centimeters to 30.5 centimeters and in weight from a few grams to 22.7 kilograms. Speeds from 12 000 to 50 000 rpm have been maintained for hundreds of hours in both air and vacuum environments. Generally, these bearings have operated satisfactorily and retained their lubricant.

TM X-53872

September 8, 1969

Terrestrial Environment (climatic) Criteria Guidelines for Use in Space Vehicle Development, 1969 Revision. Glenn E. Daniels, ed. Aero-Astro dynamics Laboratory. N70-16998.

Guidelines on probable climatic extremes and probabilities-of-occurrence of terrestrial environment data specifically applicable for NASA space vehicles and associated equipment development are presented. The geographic areas encompassed are the Eastern Test Range (Cape Kennedy, Florida); Huntsville, Alabama; New Orleans, Louisiana; the Western Test Range (Point Arguello, California); Sacramento, California; Wallops Test Range (Wallops Island, Virginia); White Sands Missile Range, New Mexico; and intermediate transportation areas. In addition, sections have been included to provide information on the general distribution of natural environment extremes in the United States (excluding Alaska and Hawaii), cloud cover, and some worldwide climatic extremes.

TM X-53873

October 1, 1969

Optimal Transfer Trajectories From Earth Parking Orbit to Various Terminal Conic Constraints and Modifications to the ROBOT Computer Program. Rodney Bradford. Program Development. N70-15035.

Reference optimal (minimum fuel consumption) transfer trajectories were computed by the minimum Hamiltonian-steepest ascent method for departure from a circular parking orbit about the earth to various terminal conic constraints. Terminal conditions were a circular orbit at synchronous altitude, 35 862 km, an elliptical orbit with apogee at synchronous altitude (in-and-out-of-parking-orbit-plane), and an injection burn into a typical 1973 Mars mission hyperbolic orbit. The vehicles used to perform these transfers were the S-IVB stage and the Command Service Module. Also included is a typical ascent trajectory profile utilizing a two-stage Saturn IB launch vehicle. Detailed orbital and trajectory profile parameters are presented in tabular form.

TM X-53874

September 9, 1969

Infrared Testing of Electronic Components. E. G. Osburn. Quality and Reliability Assurance Laboratory. Converted from IN-R-QUAL-67-2.

NASA TECHNICAL MEMORANDA

<p>TM X-53875 September 9, 1969</p> <p>Determination of Mass Characteristics S-IC-2. T. A. Henegar. Quality and Reliability Assurance Laboratory. Converted from IN-R-QUAL-67-3.</p>	<p>TM X-53882 September 12, 1969</p> <p>Titanium 8A1-1Mo-1V Crossbeam Repair. Ronald L. Nichols. Manufacturing Engineering Laboratory. Converted from R-ME-IN-67-3.</p>
<p>TM X-53876 September 9, 1969</p> <p>Parallel Gap Welding. A. M. Pasciak. Quality and Reliability Assurance Laboratory. Converted from IN-R-QUAL-67-6.</p>	<p>TM X-53883 September 12, 1969</p> <p>Space Tool Power Source Investigation. Manufacturing Research Technology Division. Manufacturing Engineering Laboratory. Converted from R-ME-IN-67-4.</p>
<p>TM X-53877 September 9, 1969</p> <p>Evaluation of Postsoldering Cleaning Agents. F. C. Osemlak. Quality and Reliability Assurance Laboratory. Converted from TM X-53877.</p>	<p>TM X-53884 September 12, 1969</p> <p>Sterilization of Unmanned Planetary Spacecraft: A Report on Current Technology. F. J. Beyerle and E. B. Snow. Manufacturing Engineering Laboratory. Converted from R-ME-IN-67-5.</p>
<p>TM X-53878 September 9, 1969</p> <p>Final Report: Leak Detector Development (contract NAS8-11898). Quality and Reliability Assurance Laboratory. Converted from IN-R-QUAL-67-9.</p>	<p>TM X-53885 September 12, 1969</p> <p>Effect of Current Cleaning Procedures on Sterilization of Spacecraft Components. Manufacturing Research and Technology Division. Manufacturing Engineering Laboratory. Converted from R-ME-IN-67-7.</p>
<p>TM X-53879 September 9, 1969</p> <p>Evaluation Testing of Thermofit Solder Sleeves. Quality and Reliability Assurance Laboratory. Converted from IN-R-QUAL-67-10.</p>	<p>TM X-53886 September 12, 1969</p> <p>Reaction of Encapsulating Foams to Specific Environmental Conditions. H. M. Walker. Manufacturing Engineering Laboratory. Converted from R-ME-IN-67-9.</p>
<p>TM X-53880 September 9, 1969</p> <p>Evaluation of Radiographic Flaw Detection in 2219 Aluminum Tig Welds. Tom Goldsberry and Jerry Barnes. Quality and Reliability Assurance Laboratory. Converted from IN-R-QUAL-67-19.</p>	<p>TM X-53887 September 12, 1969</p> <p>Simulation Test Report for Mass Transfer of Stowage Boxes Through the Airlock Module. Manufacturing Research Technology Division. Manufacturing Engineering Laboratory. Converted from IN-R-ME-67-11.</p>
<p>TM X-53881 September 12, 1969</p> <p>Monitoring Airborne Particulate Contamination. T. W. Lewis. Manufacturing Engineering Laboratory. Converted from R-ME-IN-67-1.</p>	<p>TM X-53888 September 12, 1969</p> <p>Simulation Test Report for the Multiple Docking Adapter Package-Fastening Devices. Manufacturing Research and Technology Division. Manufacturing and Engineering Laboratory. Converted from IN-R-ME-67-12.</p>

NASA TECHNICAL MEMORANDA

TM X-53889	September 12, 1969	TM X-53896	September 12, 1969
Space Tool Power Source Safety and Reliability Investigation. Manufacturing Research and Technology Division. Manufacturing Engineering Laboratory. Converted from IN-R-ME-68-3.		A Study on the Use of Fighter Aircraft To Provide a Zero g Environment in Support of Space Manufacturing Experiments. Space Simulation and Experiments Office. Manufacturing Engineering Laboratory. Converted from R-ME-IN-68-19.	
TM X-53890	September 12, 1969	TM X-53897	September 12, 1969
Roll Diffusion Bonding Development. Carl M. Wood. Manufacturing Engineering Laboratory. Converted from IN-ME-68-4.		Development of Methods for Application of Polyurethane Spray Foam Insulation Systems to Liquid Hydrogen Tanks. James M. Carter. Manufacturing Engineering Laboratory. Converted from IN-ME-69-3.	
TM X-53891	September 12, 1969	TM X-53898	September 11, 1969
Evaluation of Alcohol Sporulation Method. Manufacturing Research and Technology Division. Manufacturing Engineering Laboratory. Converted from IN-R-ME-68-5.		A New Interpretation of the Explorer XVII and Explorer XXXII Ionization Gauge Data and Its Implications to Orbital Analysis. James O. Ballance. Aero-Astrodynamic Laboratory. N69-38272.	
TM X-53892	September 12, 1969	The response of the modified Redhead magnetron density gauges used on the Explorer XVII and Explorer XXXII satellites has been analyzed by using the transmission probabilities and time response characteristics caused by geometry and gas-molecule/surface interactions. It was found that the very evident fluid dynamic response characteristics of the data required an interpretation which uses low energy and momentum exchange for the gas on the initial wall collisions with total loss of atomic oxygen. An additional signal caused by a process — most likely to be recombination — taking place on the wall of the gauge is suggested. For the example used, there is a factor of two between the derived density based on the original assumptions and the density based on the new interpretation.	
Development and Testing of the Lunar Gravity and Earth Orbital Simulator (Parallelogram). H. T. Blaise. Manufacturing Engineering Laboratory. Converted from IN-ME-68-13.			
TM X-53893	September 12, 1969	TM X-53899	September 11, 1969
Evaluation of Installation Variables for Titanium 6Al-4V Hi-Loks and Six-Wing Bolts Assembled in Titanium 8Al-1Mo-1V Sheet Material. Carl M. Wood. Manufacturing Engineering Laboratory. Converted from IN-ME-68-15.		Results of an Experimental Turbulent Boundary Layer Control Investigation. William W. Clever, II. Aero-Astrodynamic Laboratory. N70-11894.	
TM X-53894	September 12, 1969	Results of a wind tunnel test employing wall rougheners as a means of turbulent boundary layer control are presented. This method is demonstrated	
Simulation Test Report for MDA External Mobility Aids Evaluation. Space Simulation and Experiments Office. Manufacturing Engineering Laboratory. Converted from R-ME-IN-68-17.			
TM X-53895	September 12, 1969		
Simulation Test Report for Internal Mobility Aids Evaluation. Space Simulation and Experiments Office. Manufacturing Engineering Laboratory. Converted from R-ME-IN-68-18.			

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as being simple, efficient, and capable of yielding predictable results. Boundary layer thickness increases on the order of 100 percent are shown to be possible without undue flow disturbance and velocity profile distortion. The results are compared with the theoretical growth of a turbulent boundary layer over a flat plate.

TM X-53900 September 12, 1969

Trajectory Application Method (TAM). John P. Sheats. Aero-Astroynamics Laboratory. Converted from R-AERO-IN-1-67.

TM X-53901 September 12, 1969

Alternate Apollo Missions -- Libration Points. John T. Wheeler. Aero-Astroynamics Laboratory. Converted from R-AERO-IN-2-67.

TM X-53902 September 12, 1969

Derivation of Twelve-by-Twelve Stiffness Matrix for Shear Panel Undergoing Parabolic Deformation. Gary Muller. Aero-Astroynamics Laboratory. Converted from R-AERO-IN-3-67.

TM X-53903 September 12, 1969

The Use of Statistics in Guidance Analysis. Donald H. Galli. Aero-Astroynamics Laboratory. Converted from IN-AERO-68-2.

TM X-53904 September 12, 1969

Impulsive Velocity Requirements for Insertion Into Orbits About Venus, Mars, and Jupiter. Richard Gold. Aero-Astroynamics Laboratory. Converted from IN-AERO-68-3.

TM X-53905 September 12, 1969

On the Real Singularities of the N-Body Problem. Hans J. Sperling. Aero-Astroynamics Laboratory. Converted from IN-AERO-68-6.

TM X-53906 September 12, 1969

Discussion of Manual Control Problems. James H. Golmon. Aero-Astroynamics Laboratory. Converted from IN-AERO-68-7.

TM X-53907 September 12, 1969

Aerodynamic Design and Calibration of the MSFC Thermal-Acoustic Jet Facility -- Cold Flow Duct. K. D. Johnston and W. C. Tidmore. Aero-Astroynamics Laboratory. Converted from IN-AERO-69-1.

TM X-53908 September 12, 1969

Preliminary Design Program for Advanced Planetary Spacecraft. J. T. Wheeler. Aero-Astroynamics Laboratory. Converted from IN-AERO-3-69.

TM X-53909 September 12, 1969

Product Mean Values and Convection Speed. W. H. Heybey. Aero-Astroynamics Laboratory. Converted from IN-AERO-67-8.

TM X-53910 September 12, 1969

Stress Corrosion Cracking Evaluation of Several Precipitation Hardening Stainless Steels. T. S. Humphries and E. E. Nelson. Astronautics Laboratory. N69-38232.

The stress corrosion cracking resistance of precipitation hardening stainless steels PH 13-8 Mo, PH 14-8 Mo, 15-5 PH, PH 15-7 Mo, 17-4 PH, 17-7 PH, A-286, Almar 362, AM-350, and Unitemp 212 are presented. Three types of specimens (round tensile, flat tensile, and C-rings) were employed and were subjected to alternate immersion in a 3.5-percent salt solution. The results indicate that under these test conditions, PH 13-8 Mo, PH 14-8 Mo, 15-5 PH, 17-4 PH, A-286, Almar 362, and Unitemp 212 stainless steels are highly resistant to stress corrosion cracking in practically all forms and heat treat conditions.

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TM X-53911	September 16, 1969	TM X-53918	September 16, 1969
Use of Low Constant Thrust To Accomplish Orbital Transfer. Quintin Peasley and Lester Katz. Space Sciences Laboratory. Converted from IN-SSL-N-68-8.		A Numerical Integration Scheme To Determine Hemispheric Emittance, Solar Absorptance and Earth Infrared Absorptance From Spectral Reflectance Data. Donald R. Wilkes. Space Sciences Laboratory. Converted from IN-SSL-T-68-10.	
TM X-53912	September 16, 1969	TM X-53919	September 16, 1969
Electron Transport From a Cosine Law Source. Charles E. Wuller, Jr. Space Sciences Laboratory. Converted from IN-SSL-N-68-13.		Thermal Conductivity of a Particulate Sample in an Environment That Simulates the Planet Mars. James A. Fountain and Ronald W. Scott. Space Sciences Laboratory. Converted from IN-SSL-T-68-15.	
TM X-53913	September 16, 1969	TM X-53920	September 16, 1969
Statistical Theory of Inelastic Neutron Energy Spectra. B. J. Henderson and C. E. Wuller, Jr. Space Sciences Laboratory. Converted from IN-SSL-N-68-14.		A General Survey of Classical Cepheid Variable Stars. Robert M. Wilson. Space Sciences Laboratory. Converted from IN-SSL-T-68-16.	
TM X-53914	September 16, 1969	TM X-53921	September 16, 1969
The Gravitational Red Shift. Peter Eby. Space Sciences Laboratory. Converted from IN-SSL-N-68-19.		A Method for Simulating Van Allen Belt Proton Energy Spectra. John W. Watts, Jr. and Martin O. Burrell. Space Sciences Laboratory. Converted from IN-SSL-68-18.	
TM X-53915	September 16, 1969	TM X-53922	September 16, 1969
A Brief Discussion of Berger's Latest Monte Carlo Program. J. H. Derrickson. Space Sciences Laboratory. Converted from IN-SSL-P-68-12.		Film Degradation Resulting from Magnetically Trapped Protons in ATM Orbits. Space Sciences Laboratory. Converted from IN-SSL-N-69-2.	
TM X-53916	September 16, 1969	TM X-53923	September 16, 1969
Solar Flow Parameters. Klaus Schocken. Space Sciences Laboratory. Converted from IN-SSL-T-68-4.		Relativistic Time Dilation on Lunar Flights. Fred Wills. Space Sciences Laboratory. Converted from IN-SSL-N-69-4.	
TM X-53917	September 16, 1969	TM X-53924	September 16, 1969
A Microscopic Study of Solid/Liquid Phase Change in Several Members of the Paraffin Family. Barbara E. Richard and Tommy C. Bannister. Space Sciences Laboratory. Converted from IN-SSL-T-68-9.		Determining the Orientation of the Pegasus Satellites. Robert L. Holland, Fred A. Rodrigue, and Matthew B. Barkley, Jr. Space Sciences Laboratory. Converted from R-RP-INP-67-2.	

NASA TECHNICAL MEMORANDA

<p>TM X-53925 September 16, 1969</p> <p>The Thermal Structure of the Sun. Klaus Schocken. Space Sciences Laboratory. Converted from R-RP-INT-67-3.</p>	<p>TM X-53932 September 16, 1969</p> <p>Physics of the Clock Experiment. Russell D. Shelton, Nat Edmonson, Jr., and Fred Wills. Space Sciences Laboratory. Converted from R-SSL-INN-67-12.</p>
<p>TM X-53926 September 16, 1969</p> <p>An Application of Collisionless Ion Acoustic Wave Phenomena to Satellite Experiments. Lawrence H. Wood. Space Sciences Laboratory. Converted from R-RP-INN-67-4.</p>	<p>TM X-53933 September 16, 1969</p> <p>The Magnetic Field of the Sun and the Generation of Sunspots. Klaus Schocken. Space Sciences Laboratory. Converted from R-SSL-INT-67-13.</p>
<p>TM X-53927 September 16, 1969</p> <p>Ion-and Diffusion-Pump High Vacuum Systems. Philip W. Tashbar et al. Space Sciences Laboratory. Converted from R-RP-INP-67-6.</p>	<p>TM X-53934 September 16, 1969</p> <p>Coordinate Systems in Relativistic Frequency Shifts of Atomic Clocks. Fred D. Wills. Space Sciences Laboratory. Converted from IN-SSL-N-67-14.</p>
<p>TM X-53928 September 16, 1969</p> <p>Heat Transfer in the Convective Envelope of the Sun. Klaus Schocken. Space Sciences Laboratory. Converted from R-SSL-INT-67-8.</p>	<p>TM X-53935 September 16, 1969</p> <p>An Analysis of Cassegrain Optics. Joe Michlovic. Space Sciences Laboratory. Converted from IN-SSL-T-67-16.</p>
<p>TM X-53929 September 16, 1969</p> <p>Analysis of the Effect of High Energy Protons on ATM Photographic Film. Russell D. Shelton. Space Sciences Laboratory. Converted from R-SSL-INN-67-9.</p>	<p>TM X-53936 October 10, 1969</p> <p>Bolt Torque Tests in Vacuum. Keith E. Demorest. Astronautics Laboratory. N70-17607.</p> <p>A report on test equipment and tests made to determine the effects of a vacuum environment on the release torque of various threaded fastener materials both lubricated and unlubricated is given. Results show the lowest release torques were obtained with steel bolts using an inorganically bonded dry film lubricant in vacuum while the highest release torques occurred with unlubricated steel fasteners in air.</p>
<p>TM X-53930 September 16, 1969</p> <p>The Heat Pipe Experiment. R. D. Shelton. Space Sciences Laboratory. Converted from R-SSL-INN-67-10.</p>	<p>TM X-53937 June 29, 1970</p> <p>Effects of Sputtering Parameters on Teflon Thin Film Capacitors. R. I. Miller and R. C. Ruff. Astronautics Laboratory.</p>
<p>TM X-53931 September 16, 1969</p> <p>Comparison of Experimental and Theoretical Brightness Temperatures of the Lunar Surface for Different Elevation Angles of Sun and Observer. James K. Harrison. Space Sciences Laboratory. Converted from R-SSL-INT-67-11.</p>	

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The effects of varying the radio frequency (rf) sputtering parameters on the dielectric properties of resultant deposited thin films of Teflon is discussed. Electrode voltage, electrode spacing, and sputtering pressure were the parameters studied, and film thickness was the controlled variable. The dielectric constant, dissipation factor, dielectric strength, and resistivity were determined. In general, the dissipation factor increases with film thickness, while resistivity and dielectric strength decrease with increasing thickness. All parameters measured indicate a strong anomaly at 3 000-Å film thickness.

TM X-53939

May 19, 1970

Radiographic Interpretation Guide for Aluminum Alloy Welds. P. C. Duren and E. R. Risch. Astronautics Laboratory.

The problem of radiographic interpretation has plagued the X-ray method of nondestructive testing since its beginning. In an attempt to achieve a more accurate interpretation of radiographic images of aluminum alloy welds, a program designed to correlate specific images with corresponding discontinuities was initiated. Radiographs were made of welds that contained different types of discontinuities. Subsequent metallographic studies were made to confirm the image types and shapes as depicted in the radiographs. These experiments required very precise welding techniques to produce the various types of discontinuities found in typical aluminum welds. Exact radiographic techniques and careful metallographic methods were also necessary to achieve program objectives. Lack of agreement on terminology has also plagued the advancement of radiographic interpretation. Therefore, the radiographic terms used in this report have been carefully defined to eliminate as many interpretational errors as possible. The resultant data show clearly that radiographic interpretation can be placed on a scientific basis.

TM X-53940

October 10, 1969

The Nondestructive Evaluation of Low Density Foam-Aluminum Composite Materials. W. N. Clotfelter, B. F. Bankston, P. C. Duren. Astronautics Laboratory. N70-17360.

NOPCO BX-250, a polyurethane spray foam is now used as cryogenic insulation for S-II stages of the Saturn vehicle. This application has required considerable effort in the development of nondestructive methods to evaluate the mechanical integrity of foam to metal bonds. Certain inspection problems associated with low density foam-aluminum composites are evaluated in this report. The development of audio frequency methods required to overcome these difficulties are described. Additional effort was required to adapt one of these audio frequency methods and to develop radiographic techniques for the detection of voids in the foam.

TM X-53941

October 17, 1969

Spectral Emission Properties of Selected Igneous Rocks in the 2.8- to 13.5-Micron Region. William D. Hunton. Space Sciences Laboratory. X70-13226.

Several laboratories are performing studies in the area of remote identification of rocks and minerals from their infrared reflections and emission spectra. The results of the studies should be useful in such programs as Apollo Applications. However, more laboratory data on known specimens are needed to identify rocks and minerals in situ.

In this report emission properties of selected igneous rocks in the 2.8- to 13.5-micron region are discussed. A circular variable filter wheel infrared spectroradiometer was used to obtain spectra from rough rock chunks heated to between 46° and 47° Celsius. The samples showed either several shallow emission minima or just a general departure from blackbody radiation. None showed any narrow bandwidth emission minima. This work has shown that the circular variable filter spectroradiometer is a useful tool in the field of identification of rocks whose temperatures are not highly elevated. It is useful in either distant or close scanning of rocks of unknown composition. Also, the spectra included should provide a useful catalog for work in infrared emission analysis.

TM X-53942

August 22, 1969

Nucleate Pool Boiling of Saturated Freon 113 in a Reduced Gravity Environment. Jerrol Wayne Littles. Astronautics Laboratory. N69-38608.

NASA TECHNICAL MEMORANDA

Pool boiling of Freon 113 was investigated in a body force field less than standard gravity using the 4-sec drop tower. Emphasis was placed on the behavior of the nucleate portion of the pool boiling curve. The growth and departure of individual bubbles were also investigated. A transient calorimeter technique was used to obtain boiling curve information. This technique allowed the use of a heater of sufficient size to eliminate the problem of heater geometry influencing the boiling behavior while avoiding the necessity of achieving steady state during the short drop time available. The results indicate that the location of the pool boiling curve is a function of both acceleration level and surface orientation.

TM X-53943 August 29, 1969

Celestial Coordinate Transformations.
Nadine A. Bicket and Gilmer A. Gary. Space Sciences Laboratory. N70-12258.

Three FORTRAN programs required to give the coordinate transformations among the equatorial, ecliptic, and galactic systems are described. A brief review of spherical trigonometry is included. From the equations given for coordinate transformations and a development of the equatorial and ecliptic systems, a method for the general transformations was derived. These coordinate transformations were developed and programmed for use in the determination of the ecliptic (Zodiacal Light) with respect to the other systems for analysis of the S-073/T-027 AAP experiment data analysis.

TM X-53944 September 26, 1969

Design Sensitivities of Multistage Launch Vehicles. Dietrich W. Fellenz. Advanced Systems Office. Converted from IN-ASO-67-1.

TM X-53945 September 26, 1969

Parametric Sensitivity Analysis for Staged Propulsive Vehicles. G. R. Woodcock. Advanced Systems Office. Converted from IN-ASO-67-2.

TM X-53946 September 26, 1969

Staged Convoy Concepts in Lunar and Planetary Surface Exploration. Georg von Tiesenhausen. Advanced Systems Office. Converted from IN-ASO-67-4.

TM X-53947 September 26, 1970

Symmetric Round Trip Flybys to Outer Planets. G. R. Woodcock. Advanced Systems Office. Converted from IN-ASO-67-6.

TM X-53948 September 26, 1969

On Analysis of Area Coverage by Orbital Photoimaging Systems. G. R. Woodcock. Advanced Systems Office. Converted from IN-ASO-68-1.

TM X-53949 September 26, 1969

Preliminary Analysis of a Wake Trailing a Spacecraft. Walter H. Stafford and John P. Butcher. Advanced Systems Office. Converted from IN-ASO-68-3.

TM X-53950 September 26, 1969

A Graphical Approach to Convolution. James C. Taylor. Astrionics Laboratory. Converted from IN-ASTR-R-69-1.

TM X-53951 September 29, 1969

Mie Scattering: A Computer Program and an Atlas. Nadine A. Bicket and Gilmer A. Gary. Space Sciences Laboratory. N70-17291.

A brief background of Mie Scattering is given, with applications in distributions of particles and effects of different lens apertures of the flux-collecting device. A computer program, MIESCA, which calculates amplitude functions, efficiency factors, intensities, and polarizations, is listed, with adaptations for tape filing in the Univac 1108 and for plotting intensities and polarizations versus scattering angle on the SC 4020 plotter. An appendix on several numerical integrations and an atlas of scattering intensity graphs are included.

NASA TECHNICAL MEMORANDA

TM X-53952

October 21, 1969

A Stellar X-Ray Astronomy Summary and Bibliography. Robert M. Wilson, John M. Reynolds, and Stanley A. Fields. Space Sciences Laboratory. X70-12611.

This report represents a review of stellar X-ray astronomy. Tables and diagrams summarizing some of the results of the survey are shown. The right ascensions and declinations with special notes are presented for 57 X-ray sources. A tabulation of all X-ray flight experiments from the literature surveyed is shown. The report is of the annotated bibliographic, cross-referencing style.

TM X-53953

September 30, 1969

A Simplified Dead Reckoning Navigation System for the Manned Lunar Roving Vehicle. Walter L. Green. Astrionics Laboratory. N70-20655.

A dead reckoning navigation system for the manned lunar roving vehicle is presented. The basic hardware for the system is two free gyros mounted to the frame of the vehicle, the wheel odometers and tachometers, and a signal processor. The mathematical equations for the navigation system are developed and related directly to the vehicle velocity and the gimbal angle readings. A digital simulation program is written for the navigation system for performing error analysis. The simulation can be used for determining the effects of gyro drifts and system misalignments for any selected lunar surface profile. The results of a study with selected ranges of gyro drift rates and misalignments for a particular surface profile are given.

TM X-53954

August 22, 1969

Electron Bremsstrahlung Shielding at Synchronous Altitude by Electron Trapping in Dielectrics. D. L. Hollis. Space Sciences Laboratory. N70-15058.

A theoretical study is made of synchronous altitude electron bremsstrahlung shielding by electron trapping in a layer of dielectric material on the outer surface of a spacecraft. Traps in the insulating material retain bombarding electrons and thereby produce an internal spacecharge field of large

magnitude. The electric field repels or slows additional electrons. This acts to reduce bremsstrahlung production, lower the photon energy, and enhance photon absorption. The breakdown phenomenon causes the shielding effect of the trapped electrons to be cyclic. Qualitative calculations indicate that a 40-percent decrease in mass results when a thin layer of polyethylene instead of lead is used in reducing bremsstrahlung in an aluminum wall to the same level. Suggestions are given to modify existing Monte Carlo codes for this particular application.

TM X-53955

1969

Meteoroid Physics Research at MSFC. Research Achievements Review. Volume III, No. 8. Science and Engineering.

TM X-53956

October 9, 1969

Cape Kennedy Wind Component Statistics Monthly and Annual Reference Periods for All Flight Azimuths From 0- to 70-km Altitude. S. C. Brown. Aero-Astrodynamic Laboratory. N70-15048

Head-, tail-, and cross-wind component speeds for Cape Kennedy are tabulated for all flight azimuths for altitudes from 0- to 70-km by monthly and annual reference periods. Wind speeds are given for 11 selected percentiles ranging from 0.135 to 99.865 percent for each reference period. The data were prepared for application to space vehicle dispersion analysis and pitch (or yaw) program bias to minimize vehicle structural and control loads caused by wind.

TM X-53957

October 17, 1969

Space Environment Criteria Guidelines for Use in Space Vehicle Development (1969 Revision). Don K. Weidner, ed. Aero-Astrodynamic Laboratory. N70-22208.

A cross section of natural environment data for use as design criteria guidelines in space vehicle development programs is presented. Specifically, information is provided in the disciplinary areas of atmospheric and ionospheric properties, radiation, solar cycle predictions, geomagnetic field, astrodynamical constants, and meteoroids for the earth's atmosphere above 90 kilometers, interplanetary space, and the atmospheres and surfaces (when available)

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of the moon and the planets (other than earth) of this solar system. The current MSFC upper atmosphere model and solar cycle prediction routines are described in detail.

TM X-53958

October 14, 1969

Large Angle Method for Space Vehicle Angular Momentum Desaturation Using Gravity Gradient Torque. Hans F. Kennel. Astrionics Laboratory. N70-19867.

The angular momentum desaturation method is based on the use of the gravity gradient torques. Large angular maneuvers maximize the effectiveness of the desaturation method for the case where the mission requires the minimum moment-of-inertia axis to point toward some celestial object for the nonocculted part of the orbit. The desaturation method is developed under the assumption that the difference between two of the principal moments of inertia is negligible. A constant attitude with respect to the gravity gradient direction is held during the desaturation period. This allows maximization of the available gravity gradient desaturation torque which is necessary since the assumed moment-of-inertia distribution (in relation to the mission requirements) is a very poor candidate for angular momentum desaturation using the gravity gradient, especially when relatively large aerodynamic disturbance torques are present.

TM X-53961

November 21, 1969

Holographic Optical Data Processing. Rodney W. Jenkins. Space Sciences Laboratory. N70-17359.

To process optical data is to perform mathematical operations on optical inputs. This paper describes the holographic technique and its application to optical subtraction and pattern recognition (matched filtering). Also, experimental results that are presented verify the theory, and the appendixes show the Fourier transformation operator property of a lens and input translation invariance of the matched filtered output.

TM X-53962

1969

Computation Research at MSFC. Research Achievements Review. Volume III, No. 9. Science and Engineering.

TM X-53963

March 3, 1970

Investigation of the Development of Cracks in Solder Joints. Barnes Beasley. Astrionics Laboratory.

A study was made to investigate the development of cracks in solder joints on printed circuit boards and to recommend solutions to this problem. This study consisted of two approaches. One approach, an analytical investigation, was to perform a mathematical analysis of the joints in question with emphasis on the force applied to the solder joint; the second approach, an empirical investigation, was performed to determine the extent of damage caused by temperature cycling of printed circuit boards and to determine or evaluate the practical validity of the proposed solutions.

TM X-53965

December 12, 1969

Communication for the 70's. Scientific and Technical Information Division. Management Services Office.

These presentations were given at the Ninth Annual Society of Technical Writers and Publishers (STWP) and Association of Technical Artists (ATA) Seminar held in Huntsville, Alabama from October 21 to October 23, 1969. The report emphasizes the vital role of effective communication within technical fields. Scientific knowledge was disseminated through various media: reports, illustrations, presentations, and simple word of mouth.

TM X-53966

November 17, 1969

ARC Guidance Development Work for Tack-Welded Joints. James Chadbourne and W. A. Wall. N70-10699.

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In an attempt to leap-frog long tack welds, a reliable tack weld detector was developed to sense approaching tack welds. This sensor is perhaps the most significant development to result from the phase of the program. It is concluded that the present MSFC system should not be located close to the torch when tack welds are employed to hold the weld joint; however, it is recommended that the present electro-inductive system should be considered a boon for the development of an automatic tack welder because of its homing capabilities. With a memory system, it seems feasible that the MSFC developed system could be used with short tack welds. As an evaluation and a comparison of MSFC's transducer with the latest optical tracking transducer is drawn to emphasize the strengths and weaknesses of each transducer.

TM X-53967

November 17, 1969

An Evaluation of Termi-Point Connectors. Quality and Reliability Assurance Laboratory. (Converted from IN-R-QUAL-66-28). N70-10500.

This report is a statistical evaluation of termi-point connectors and their adequacy for use in ground support equipment. Tests performed included vibration, physical shock, salt spray, gas seal, thermal shock, and humidity. Suitability for the intended purpose was demonstrated; however, it was felt that in certain critical areas and for flight conditions, a more dynamic test program should be conducted.

TM X-53968

November 17, 1969

Evaluation Testing of Thermofit Solder Sleeves. Quality and Reliability Assurance Laboratory. (Converted from IN-R-QUAL-67-10). N70-10599.

Data obtained during evaluation testing of thermofit solder sleeve D-101-31 when used on nickel clad copper wire and alloy 63 wire is discussed. The tests performed include voltage drop, peel strength, dielectric strength, water immersion, moisture resistance, high temperature aging and vibration. A test was conducted to determine the optimum heating time of the solder sleeve to obtain maximum peel strength of the joint. Another test, using additional flux was performed to obtain better wetting of the shield. It was found that very good wetting of the connection was obtained when the shield braid was

prefluxed; however, tests indicated that a corrosive flux residue remained in the connection. Voltage drop and peel strength tests of stub splices, a combination of crimp ferrules and solder sleeves, were included.

TM X-53970

December 23, 1969

Development of Polymeric Fuel Tank Sealants for Advanced Aerospace Vehicles. William J. Patterson and Lawrence R. Moffett, Jr. Astronautics Laboratory. N70-17418.

This report describes studies in the development of a polymeric material capable of functioning as a fuel tank sealant in hypersonic aircraft. The preparation of a number of organic intermediates is outlined which encompasses a variety of systems including aryl-fluorinated compounds and aromatic polycyclic structures. Preliminary results indicate that incorporation of fluorine atoms into the basic polymer structure imparts a degree of aviation fuel resistance, a property which the sealant must of necessity possess. The rationale supporting the selected polymeric systems chosen for study and investigation is outlined in detail along with other pertinent factors involved in the selection of candidate materials. Supplemental studies involving the development of thermally resistant polymers as fuel tank sealants are described.

TM X-53971

November 19, 1969

Stress Corrosion Cracking of Ti-6Al-4V Titanium Alloy in Various Fluids. James G. Williamson. Astronautics Laboratory. Astronautics Laboratory. N70-15043.

An investigation of the stress corrosion susceptibility of Ti-6Al-4V alloy in various fluids was initiated because of the recent unexpected failures of this alloy in both nitrogen tetroxide and absolute methanol. This study has indicated that Ti-6Al-4V alloy is susceptible to stress corrosion cracking in absolute methanol in both the annealed and solution treated and aged (STA) conditions. The threshold value for annealed material appears to be near 50 percent of the yield strength or approximately 70,000 psi, while the value of STA material is somewhat less. Long term exposure tests (two years) indicated that the Ti-6Al-4V alloy is not susceptible to stress corrosion cracking in acetone, Aerozine 50, distilled

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water, absolute ethanol, Freon PCA, Freon MF, hydraulic oil (MIL-H-5606), absolute isopropanol, monoethyl hydrazine, RP-1 fuel and trichloroethylene.

TM X-53972

November 25, 1969

Traction Drive System Design Considerations for a Lunar Roving Vehicle. Clyde S. Jones, Jr., Billy J. Doran, and Frank J. Nola. Astrionics Laboratory. N70-18797.

For an optimum design, the weight, energy consumption, and operational flexibility of the traction drive system for a lunar roving vehicle must be considered along with the power supply, motor, and power train. Other problems considered in this paper include: environment and motor dissipation, motor type (i.e., ac or dc) and commutation if dc, motor controller (i.e., switching of large currents), delivery of torque at varying speeds, power train, use of regenerative braking and conservation of energy, and power supply voltage variation. These problems are studied in view of certain general system specifications, which fall into weight, performance, and environment categories. Tradeoff studies are considered for optimization in each of these areas. Special consideration is given to the controller and system design as it pertains to regenerative braking and the conservation of energy. Some aspects of changing the modes of motor operation to satisfy extreme performance requirements of speed and torque are discussed.

TM X-53973

June 30, 1970

Space Flight Evolution. Georg von Tiesenhausen and Terry H. Sharpe. Program Development.

This report describes a possible comprehensive path of future space flight evolution. The material in part originated from earlier NASA efforts to define a space program in which earth orbital, lunar, and planetary programs are integrated. The material presented is not related to specific time schedules but provides an evolutionary sequence. The concepts of commonality of hardware and reusability of systems are introduced as keys to a low cost approach to space flight.

TM X-53974

January 5, 1970

The Characteristics of Penetration for a Double-Sheet Structure With Honeycomb. David W. Jex, Archie M. Miller, and Charles A. MacKay. Space Sciences Laboratory. N70-25272.

The channeling of debris by a double-sheet structure separated by a honeycomb material has been reported by many investigators in the hypervelocity field. It has been suggested that this channeling will reduce the protective capability of such a structure when compared to an identical structure without honeycomb. A series of tests was conducted by this laboratory to gain an understanding of the characteristics of the channeling phenomena and the extent to which the channeling would affect the ballistic limit of a honeycombed structure. It was found that the particular honeycomb tested exhibited greater protective ability than an identical double-sheet structure without the honeycomb for velocities between 3.5 and 8.23 km/sec.

TM X-53975

January 9, 1970

Flat Conductor Cable Design, Manufacture, and Installation. W. Angele and James D. Hankins. Manufacturing Engineering Laboratory.

This report presents pertinent information for hardware selection, design, manufacture, and quality control necessary for flat-conductor cable interconnecting harness application. The relation of the hardware to the new MIL-C-55543 and MIL-C-55544 specifications for flat-conductor cable and flat-conductor cable connectors is defined. Comparisons are made between round-wire cable and flat-conductor cable. The flat-conductor cable interconnecting harness systems show major cost, weight, and space savings, plus increased system performance and reliability. The design application section includes electrical characteristics, harness design and development, and a full treatise on EMC considerations. Manufacturing and quality control sections pertain primarily to the NASA/MSFC developed conductor-contact connector system and special flat-conductor cable to round-wire cable transitions. This report presents flat-conductor cable technology as of June 1968 and predicts the role of flat-conductor cables in interconnecting harness applications for future programs.

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TM X-53976

February 16, 1970

Conceptual Design of a High Energy Astronomy Observatory. Volume I: Preliminary Analysis. Volume II: Appendices. Program Development. Program Development. N70-22901.

The preliminary analysis and conceptual design of a baseline spacecraft along with supporting technical data and discussion of mission and spacecraft alternatives are presented.

The High Energy Astronomy Observatory treated in this work is the first of four planned spacecraft in the High Energy Astronomy Observatory program, designated in this report as HEAO-A. The primary mission objective of the HEAO-A spacecraft is to completely survey the celestial sphere for high energy X-rays, gamma-rays, and cosmic-rays, with primary emphasis on the galactic belt region; the secondary mission objective is selective pointing at specific celestial targets.

To ensure a comprehensive system analysis for feasibility assessment, a baseline mission and spacecraft was defined with a realistic, but hypothetical, experiment package. Total weight of the baseline spacecraft is approximately 19 000 pounds and launch is assumed from ETR on the Titan IIID launch vehicle in March 1974. The satellite is placed into a 200-n.mi. circular orbit with a 28.5-degree inclination; during the first month in orbit the slowly rotating satellite scans a region measuring ± 8.5 degrees from the galactic plane; during the next 6 months the entire celestial sphere is scanned; and during the last 5 months of the first mission year the satellite employs a pointing mode for selected source investigations. Satellite design lifetime is 1-year minimum, with 2 years desired.

TM X-53978

January 19, 1970

Cryoformed 301 Stainless Steel for Pressure Vessels. C. N. Irvine. Manufacturing Engineering Laboratory. N70-22076.

The manufacture and qualifying results of cryogenically formed 301 stainless steel bottles for storage application, especially for helium in either a liquid oxygen or liquid hydrogen environment, are reviewed. No apparent distortion to the prototype

vessels occurred during the course of testing. Survival of the test specimens through the qualification tests indicates that the developed manufacturing technology results in predictable properties and verifies the reliability of the techniques for the proposed applications. The cryoformed 301 stainless steel vessels should be considered for future launch vehicle and space applications, especially when weight is critical or a liquid or gaseous oxygen environment may be encountered or when temperatures vary from 394.3°K (250°F) to liquid hydrogen or helium temperatures.

TM X-53979

January 5, 1970

An Improved Method for Thin-Film Thickness Measurements. James L. Zurasky and Robert Forsten. Astrionics Laboratory. N70-19806.

An improved method of measuring thin films on metal by interpretation of interference fringes is described. The essential characteristic of the method is photographing an interference pattern produced by an interferometer; then, this negative is scanned by use of a microdensitometer to determine exact fringe location. Details of the method, expected accuracies, and possible improvements are outlined.

TM X-53982

December 22, 1969

Wind Determination Based on Eddy Transit Times Measured Between Four Nonintersecting Light Beams. W. H. Heybey. Aero-Astroynamics Laboratory. N70-21988.

An atmospheric eddy crossing successively the lines of sight (beams) of two telescopes stays between them for a time that can be discovered through cross correlation of the light intensity variation recorded by the receivers. The length and location of the paths vary with the beam arrangement and with the wind vector which is taken as constant throughout a suitably bounded observation volume. At least four detectors are required for monitoring winds of unknown azimuthal direction. They are conveniently placed at the corners of a square sitting on level ground. Restraints determine the range in which an observed transit time triplet can be trusted to have been generated by one and the same wind, measured accurately enough to warrant the computation of its components from the expressions given for them.

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TM X-53983

February 9, 1970

Fast Carry Accumulator Design. William C. Mastin. Astrionics Laboratory.

Methods for increasing the speed of binary addition by decreasing carry propagation time are reviewed. Particular attention is given to those methods that would be applicable to an accumulator using ones complements with end-around carry for subtraction. An iterative accumulator using pulse logic for input and carry signals is described. Realizations of gated carry, carry-completion detection, and carry-skip circuits that would be compatible with this accumulator are presented. NAND gates are used in the design of the required combinational networks.

TM X-53984

February 16, 1970

Calculations of Astrophysical Partition Functions. M. J. Hagyard. Space Sciences Laboratory.

A method for the calculation of the internal energy partition function of an atom or ion immersed in a plasma is presented. Sample calculations are made at various temperatures and electron pressures, and the numerical results are compared with those of other authors. The importance of the contributions of higher parent terms is clearly demonstrated.

TM X-53985

January 16, 1970

The Development of HXW Thermal Coating. Roger J. Harwell. Astronautics Laboratory. N70-18850.

The development of a coating for use on space vehicles to control their temperature while in a space environment is described. This is accomplished by varying the optical properties of the coating to control thermal absorption and emission rates of the vehicle surface. The inorganic coating described within this report has the capability of providing the necessary optical properties for thermal control and is also very stable in the environment of vacuum and solar radiation. The HXW coating is primarily a water suspension of synthetic mica that is gelled with a silicate solution. The products used in the coating are available from commercial sources and are relatively inexpensive for this type of coating.

TM X-53986

January 19, 1970

Contribution of the Spatial Part of the Metric Tensor to the Observed Frequency Shift in the Maser Clock Experiment. Peter Eby. Space Sciences Laboratory. N70-22124.

A formula for the frequency shift of a maser clock in orbit relative to an identical maser on the earth is derived. In addition to the usual terms, because of the gravitational potential difference and special relativistic effects, this formula is shown to contain terms that depend on the spatial part of the metric tensor. These extra terms cannot be derived using only the equivalence principle. It is shown that the frequency shift given by these additional terms appears to be observable with presently attainable maser accuracies. The measurement of the additional effects will yield the same information about the metric as will the geodetic precession of a gyroscope or the deflection of light.

TM X-53987

February 3, 1970

A Conceptual Design of the Space Shuttle Integrated Avionics System. Fred E. Digesu, ed. Program Development. N70-25104.

The status of an investigation to define an integrated avionics system for the Space Shuttle is presented. The system concept depicted for the integrated approach, based on subsystem requirements established early in the study, meets the mission requirements outlined in Shuttle Task Group reports and is compatible with the significant features of the vehicle configurations resulting from several 1969 contractor design studies. Included as part of the integrated avionics system is all Shuttle equipment that utilizes electronics in performance of the various functions required to accomplish the Space Shuttle mission. The individual subsystems are defined in varying degrees of detail and all data will be subject to change as the design work continues. Sufficient data are presented, however, to identify the significant physical characteristics, important performance values, functional block diagrams, and preliminary hardware selection for each avionics subsystem.

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TM X-53989

January 31, 1970

Fifty-Man Space Base Population Organization.
Georg von Tiesenhausen. Program Development.

An attempt is made to establish a baseline social and functional structure for a 50-man Space Base and to show how the requirements and activities of the personnel affect the Space Base layout.

TM X-53990

February 27, 1970

Torsional Vibration Analysis of Saturn Vehicles. Larry Kiefling and Frank Bugg. Aero-Astroynamics Laboratory.

Methods for finding the torsional vibration modes for either a beam-like vehicle, such as Saturn V, or a multibeam vehicle, such as Saturn IB, are developed. The single beam analysis is a Stodola iteration method. The Saturn IB vehicle is mathematically modeled as a system of beams and connecting members. The modes of the vehicle are composed of the superimposed components of normal beam modes plus rigid body motion. A comparison is made with dynamic test data. The calculated frequencies differed from the experimental values by an average of 8 percent for the cases compared.

TM X-53992

February 2, 1970

A Systems Engineering Decision Algorithm With Application to Apollo Applications Program Integration Problems. C. E. DeSanctis. Astronautics Laboratory. N70-20093.

A decision algorithm, that has been specifically adapted to the needs of the System Engineer, is presented for resolving complex Apollo Applications Program (AAP) technical and management integration problems. An explanation of decision theory as applied to a technical problem and a detailed example of how the decision algorithm was used to determine the best location for the scientific airlock in either the orbital assembly's airlock module structural transition section, or the multiple docking adapter is submitted. It is concluded that alternate position number 1, located on the structural transition section, be chosen to resolve the AAP technical, operational, and management requirement problems.

TM X-53993

February 5, 1970

Space Processing and Manufacturing.
Manufacturing and Engineering Laboratory.
N70-20517.

Production engineering aspects of materials processing and industrial manufacturing are discussed with application to orbiting laboratories and orbital workshops. Analyzed are especially the effects of reduced gravity or weightlessness on production processes.

TM X-53994

February 6, 1970

A Collection of Lower Thermospheric (100- to 300-km-Altitude) Chemical Composition, Temperature, and Mass Density Data.
Don K. Weidner and Michael T. Calloway.
Aero-Astroynamics Laboratory.
N70-20367.

Atmospheric data were obtained by various investigators from 38 rocket-probe flights and two satellite-borne absorption spectrometers. The data sample is by no means complete, but should provide valuable inputs to studies concerning the structure and variability of the atmosphere. Atmospheric temperature, mass density, and constituent number densities of N₂, O₂, O, He, and H as calculated from the MSFC Modified Jacchia Model Atmosphere, 1967, for the conditions of a particular observation are tabulated from 120 to 1000 km for most of the observations. These data should also provide valuable inputs to various studies. In using these data, however, the investigator should consider the limitations (constant 120-km boundary conditions, empirical temperature, etc.) of the MSFC Model Atmosphere.

TM X-53996

February 12, 1970

The Generation of X Rays in Planetary Nebulae. Klaus Schocken. Space Sciences Laboratory.

The evolution of planetary nebulae and the production of X rays will be summarized from the literature. Planetary nebulae emit continuous radiation which originates by free-free transitions involving kinetic energy losses of free electrons in

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the electrostatic fields of ions. These free-free emissions are responsible for the radiation observed in the radio frequency spectra. They also contribute in the visible and infrared regions. Therefore, if X rays are generated by planetary nebulae, it can be assumed that they will also be produced by free-free transitions. However, free-free radiation will be measured along with free-bound radiation. A formula is given for the intensity of the combined continuum for an ionized gas with a Maxwellian distribution of the electron velocities.

TM X-53998

March 13, 1970

X Radiation in Binary Stars. Klaus Schocken. Space Sciences Laboratory.

The properties of close binary stars and old novae as they are known will be summarized from the literature sources listed in the bibliography. A binary system originates with or results from the splitting of a single star into two components caused by rotational instability during the contraction of the star. The assumption is made that, to become a nova, it is a necessary condition for the star to be a member of a close binary system consisting of a blue and a red star, with a stream of gas flowing from the red to the blue star and forming a shell around the blue star to generate the observed bright emission lines that indicate that this gaseous shell moves with the blue star. Assuming that the explosive process has taken place in the nova, a shock wave will penetrate the surface, forming an expanded atmosphere in a time that is too short for appreciable loss of radiation. When an electric field exists in a plasma with a density such that the electrons whose velocity surpasses a certain critical value will gain more energy from the field than they lose by collisions, runaway electrons will be generated. The hypothesis is made that runaway electrons are the source of deceleration X radiation in the atmosphere of a nova.

TM X-64500

February 27, 1970

Space Shuttle Guidance. Wilton E. Causey. Aero-Astro dynamics Laboratory.

The guidance requirements for the Shuttle Vehicle and the possible candidate guidance schemes are discussed. The MASCOT Guidance Scheme appears to be ideally suited for the rendezvous guidance

phase of the flight, and by modifying the guidance equation to include the lift and drag forces, the MASCOT scheme can be adapted for the ascent guidance phase. The basic parts of the MASCOT guidance scheme, a numerical integration algorithm and an iteration algorithm, can be used to develop a fast-prediction guidance scheme for the reentry phase.

TM X-64501

February 27, 1970

Some Photographic Results of the Apollo 11 Mission. Paul A. Larsen. Aero-Astro dynamics Laboratory.

The work reported in this document was done for several reasons: (1) to show what types of photographs are available as a result of the Apollo 11 mission; (2) to give the best possible photographic description of the lunar terrain in the vicinity of Tranquillity Base, and as many other areas of the lunar surface as possible, so that Lunar Roving Vehicle (LRV) mission planners, designers, and engineers can obtain a "feel" for several types of lunar terrain that may be encountered by the lunar roving vehicle; (3) to show some of the better pictures as single photographs, stereograms, constructed panoramic views, sequence strips, and mosaics; (4) to demonstrate the feasibility of making certain elevation measurements of lunar topographical features, by using lunar photographs, photogrammetric techniques, and locally available photogrammetric equipment; and (5) to provide a reminder of the value of high altitude earth photography for earth resources studies.

TM X-64503

March 11, 1969

Gravity Gradient Measurements with a Laser Absolute Gravimeter. R. C. Borden and O. K. Hudson. Space Sciences Laboratory.

A technique for measuring the gravity gradient is derived from extensions of Dr. Hudson's original theory employed in the laser interferometry method for measuring absolute gravity. The performance of the proposed experiment for measuring gradient is predicted through analyzing the effects from expected sources of measurement error.

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TM X-64505

March 17, 1970

Research on Electromagnetic Correlation Techniques. Fritz R. Krause, ed. Aero-Astro dynamics Laboratory.

Advanced data processing and statistical analysis methods have been applied to output signals of active and passive electromagnetic sensors. Single path observations, intersecting beams, and separated lines of sight have been used to study the overall level, spectroscopic structure, and motion of local and remote optical disturbances. The potential of the associated "electromagnetic correlation techniques" is illustrated by model experiments in a jet, in a glow discharge, and in a blow torch. Analytical concepts for remote probing of the earth's atmosphere are illustrated through a discussion of special cross-beam arrangements for ground wind and turbulence detection. Potential applications to the general problems of separating dispersion phenomena from local photochemical processes are introduced through a piecewise accumulation of phase and gain spectra.

TM X-64506

April 17, 1970

An Approach to Contamination Identification. Dr. Thomas R. Edwards. Space Sciences Laboratory.

A regression analysis method has been applied to unfold the waveforms resulting from complex gaseous mixtures in mass spectral analysis. The output format specifies both qualitative and quantitative chemical analysis. This method of data analysis is applied towards optical contamination identification in vacuum systems. A number of test mixtures are presented to demonstrate the feasibility of the technique.

TM X-64507

February 27, 1970

An Analysis of Potential Space Shuttle Cargo-Handling Modes of Operation. Walter E. Whitacre. Program Development.

This report attempts to indicate the current status of Space Shuttle cargo-handling analysis. The problems involved in orbital docking, payload extraction and transfer, cargo handling, and special purpose missions are discussed and some tentative conclusions and recommendations are presented.

TM X-64508

April 15, 1970

Environmental Control and Life Support Subsystem (EC/LSS) for the 1975 Space Station. Hubert B. Wells. Program Development.

This report contains the results of a preliminary study to define an Environmental Control and Life Support Subsystem (EC/LSS) that is applicable to a long-term earth-orbital space station for the 1975 time period. The Space Station is capable of supporting a 12-man crew continuously over an extended period of time with regular resupply. The EC/LSS must maintain a system life requirement of 10 years through maintenance spares and redundancy. A survey was made to define a group of assemblies that is suitable for fulfilling the requirements of the EC/LSS. The primary assemblies are as follows: atmospheric supply and pressurization; oxygen recovery; atmospheric purification; thermal control; water management; water reclamation; waste management; suit loop/PLSS; crew systems; and expendables. This report contains detailed descriptions of primary assemblies, including design criteria, approaches, advantages, disadvantages, component descriptions, preliminary weight, volume, and power summaries, and other pertinent information.

TM X-64509

April 30, 1970

Experimental Development Program for Lunar Surface Navigation Equipment. Bobby F. Walls, William C. Mastin, and Peter H. Broussard, Jr. Astrionics Laboratory.

An examination of some of the proposed schemes for manned lunar surface navigation is made with overtones of unmanned navigation requirements. The examination is made with the view in mind of actual near-term implementation with available and proven hardware. An outline of MSFC's efforts in this area is given, with particular emphasis on the present, ongoing in-house design, fabrication, and testing of a candidate dead reckoning system utilizing gyro references. The ability of this system to meet the requirements of simplicity, reliability, low cost, and ruggedness is discussed.

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TM X-64510

April 2, 1970

A Stress Corrosion Evaluation of AISI-H-11 Alloy Steel Plated With Nickel-Cadmium Coating. J. W. Montano. Astronautics Laboratory.

This report presents the mechanical properties of AISI - H-11 alloy steel bar specimens (0.125-in. diameter), C-ring specimens (1.0-in. diameter) and charpy V-notched impact specimens, heat treated to 220- to 240-ksi ultimate tensile strength range and plated with nickel-cadmium. The tensile specimens and C-ring specimens were stressed to 0, 50 and 90 percent and 25, 50 and 90 percent of the 0.2-percent longitudinal yield strength respectively, and exposed to: (a) 180 days of exposure to the atmospheric conditions prevalent at the Marshall Space Flight Center during the months of December through June, and (b) 180 days of alternate immersion (A.I.) testing in a 3.5-percent sodium chloride solution. Unplated control specimens were also used in the stressed and unstressed condition for comparison purposes.

TM X-64511

April 2, 1970

Stress Corrosion Cracking Evaluation of Several Ferrous and Nickel Alloys. T. S. Humphries and E. E. Nelson. Astronautics Laboratory.

The stress corrosion cracking characteristics of H-11 steel, Hastelloy C, Inconel 718, Waspaloy, music wire, and stainless steels Arde (cryoformed) low silicon 301, AISI 303, AISI 304, Armco 21-6-9 are described. Three types of specimens (round tensile, flat tensile, and C-ring) were employed and the main test environment was alternate immersion in 3.5-percent salt solution. The results of the test indicated that music wire, Hastelloy C, Inconel 718, Waspaloy, and stainless steels Arde 301, AISI 303, AISI 304 and Armco 21-6-9 in most forms and heat treat conditions are resistant to stress corrosion cracking in environments containing moist chlorides. However, another investigator has reported both service and laboratory stress corrosion cracking of cryoformed 301 stainless steel under different conditions. H-11 steel is susceptible to stress corrosion cracking in both moist chloride environments and semi-industrial atmospheres.

TM X-64513

April 30, 1970

Atmospheric Effects on Wave Propagation at 10.6 Microns. William E. Webb, Kermit H. George, and Peter Marrero. Astrionics Laboratory.

A CO₂ laser communications system using optical heterodyne detection has been operated over a 3.2-km path at the Marshall Space Flight Center. Both scintillation and heterodyne measurements were made for varying receiver apertures and under varying atmospheric conditions. The results have been analyzed to determine the statistical distribution function for the intensity fluctuations, the atmospheric structure constant, and the power spectrum. Signal-to-noise measurements indicate that reliable heterodyne communications are possible even under deep scintillation conditions.

TM X-64514

April 8, 1970

An Optical Image Comparator for Examination of Field Ion Micrographs. J. C. Gregory and I. Dalins. Space Sciences Laboratory.

This document is a description of a device that permits direct comparison of two complex recorded images or patterns. Any small differences between the two images, normally difficult to detect by visual inspection, are distinctly illuminated and readily noticed by the observer despite the mass of complex detail common to both images. The instrument was designed as an aid to single atom studies in the field ion microscope, but may also be of application in astronomy.

TM X-64516

April 15, 1970

Investigation of the Technical Micronics Control (TMC) Process for Electropolishing Various Metals. James R. Lowery. Astronautics Laboratory.

The Technical Micronics Control (TMC) process for electropolishing a variety of metals and metal alloys was investigated. The electropolishing solution produced lustrous surfaces on stainless steel, brass, copper, nickel, aluminum, and steel; it produced only semibright surfaces on Invar and René 41, but produced a considerable amount of metal smoothing. The passivation of stainless steel was improved

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substantially by this method over that of conventional acid passivation methods. TMC polished surfaces of stainless steel, brass, and carbon steel did not accept an electroplate with good adherence but such surfaces were more resistant to tarnish or corrosion than mechanically polished surfaces. Based on results of this study, the TMC process is recommended for polishing and smoothing most metals and for passivating stainless steel.

TM X-64519

April 27, 1970

The Potential Temperature Profile in the Planetary Boundary Layer. George H. Fichtl and Julian F. Nelson. Aero-Astro dynamics Laboratory.

Nineteen observations of the temperature profile are used to analyze and to develop a model of the potential temperature profile $\bar{\theta}(z)$ in the planetary boundary layer. The observations consist of mean flow temperatures observed at the 3-, 18-, 30-, 60-, 120-, and 150-meter levels at the NASA meteorological tower site at Kennedy Space Center, Florida. It is provisionally concluded that the dimensionless potential temperature gradient $\phi_\theta = (z/T_{*0})\partial\bar{\theta}/\partial z$ is a function of z/L_0 and $R_L = -fL_0/u_{*0}$, where T_{*0} and u_{*0} denote surface values of the friction temperature and friction velocity, L_0 is the surface Monin-Obukhov stability length, and f is the Coriolis parameter. The expression $\phi_\theta = (1 - \gamma z/L_0)^{-1/2}$ summarizes the experimental results reasonably well, where γ is proportional to $R_L^{-3/2}$. The function ϕ_θ is integrated to yield the potential temperature profile and estimates of the potential temperature drop across the planetary boundary layer.

TM X-64522

May 18, 1970

Response Tests of Cup, Vane, and Propeller Wind Sensors. Dennis W. Camp, Robert E. Turner, and Luke P. Gilchrist. Aero-Astro dynamics Laboratory.

The results of a wind tunnel test program performed to investigate the response parameters of two types of cup anemometers, two types of wind

vanes, and a propeller anemometer are presented. The distance constant for the cups (wind speed sensor) was found to be approximately 0.8 and 1.3 meters for the two types tested. Values for the damping ratio of approximately 0.48 and 0.40 were found for the two types of vanes. The main point of interest in the investigation of the propeller-type anemometer, other than its response, was how the indicated speed varied as a function of angle of attack. For an angle of attack (angle formed by the longitudinal axis of the anemometer and the wind flow direction) within ± 30 degrees into or away from the flow direction, a cosine curve would give a good approximation to the data.

TM X-64523

May 20, 1970

RF Radiation Hazards to Space Station Personnel. R. A. Inman. Astrionics Laboratory.

Calculation of typical values of RF power density in the vicinity of space station antennas are included in this document. More definite calculations can be made when the system configurations have been more completely defined. However, the values calculated demonstrate that the currently accepted safe limits of RF radiation can be exceeded easily in the near field of high gain antennas on the space station. Personnel working in the field of these antennas should be protected from overexposure to this radiation.

TM X-64524

June 22, 1970

Radiation Degradation of Selected Films. William Oran. Space Sciences Laboratory.

The change in base fog caused by radiation damage was measured for the following film types: SO-180, EF 7241, EFB 7242, SO-246, SO-267, 2475, Plus-X, Tri-X, and 1N. The films were exposed to Co^{60} gamma rays in addition to 51.1- and 130.6-MeV protons. Graphs of density change versus radiation exposure are included in this report. Sensitometric curves are also given for SO-246, SO-267, 2475, 1N, Plus-X, and Tri-X films.

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TM X-64526

June 8, 1970

Solar Indices for the Period January 1, 1955
Through December 31, 1969. H. C. Euler.
Aero-Astroynamics Laboratory.

This report lists solar-geomagnetic indices from January 1, 1955 through December 31, 1969 which may be used for model atmosphere studies. These tables contain daily sunspot number, solar 10.7-cm flux and the daily geomagnetic A_p index. Also included are the three-hourly K_p and a_p indices and 27-, 54-, and 81-day running means of sunspot numbers and solar flux.

TM X-64529

June 15, 1970

Table of Values of Integrals for the Longitudinal and Lateral von Karman Turbulence Spectra. Douglas D. Mackiernan. Aero-Astroynamics Laboratory.

The longitudinal and lateral von Karman turbulence spectra are integrated over the domain $L\Omega_c < L\Omega < \infty$, where L is the integral scale of turbulence, Ω is a wave number and Ω_c is a cut-off wave number which defines the lower bound of integration. The integration procedure consists of expanding the spectra in Maclaurin series and integrating the resulting series term by term to yield tables of σ_u/σ and σ_w/σ as functions of $L\Omega_c$, where σ_u and σ_w are the contributions to the longitudinal and lateral deviations of turbulence from the Fourier components in the wave number domain $\Omega_c \leq \Omega \leq \infty$. The total longitudinal and lateral standard deviations are both equal to σ which is obtained by setting $\Omega_c = 0$. The tables provide values of σ_u/σ and σ_w/σ for $10^{-3} \leq L\Omega_c \leq 10^6$. A sample calculation is provided to aid the user.

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TN D-5206

February 1970

Turbulence Measurements in Supersonic, Shock-Free Jets by the Optical Crossed-Beam Method. M. J. Fisher and K. D. Johnston. Aero-Astroynamics Laboratory. N70-20577.

A new optical crossed-beam technique was used to measure the turbulence properties of two supersonic, shock-free, cold air jets. The method is based on the absorption of ultraviolet radiation by air. The measurements were highly successful in that reasonable and consistent values of the turbulence properties such as convection speeds, length scales, and spectra were obtained, and the data generally followed the trends set by hot-wire measurements in subsonic jets. The most interesting feature of the crossed-beam technique is the possibility of a direct measurement of three-dimensional spectral density. However, the demonstration that the crossed-beam technique can be used for turbulence measurements in supersonic flow is considered to be the major accomplishment.

TN D-5371

August 1969

Longitudinal Mass-Spring Modeling of Launch Vehicles. Rudolf F. Glaser. Astronautics Laboratory. N69-34970

To determine longitudinal axisymmetric vibration modes on launch vehicles, simplified models must be established. As a first step, a launch vehicle can be idealized by a chain of masses, helical springs, and liquid-filled containers. However, the realistic mass-spring representation of the liquid propellants supported by the elastic container wall is the problem. In this report, application of Galerkin's method to the equations of motion of the above simplified system is proposed, using the container-liquid modes with special normalization factors as coordinate functions. This method, identical with the Ritz method in this case, is favorable in several aspects. First, the resulting model is based on the energy law. The use of the container-liquid modes, however, results in accurate representation of significant structural and inertial characteristics of the container-liquid system. The special normalization of the coordinate functions converts the container-liquid systems directly in lumped parameter systems, having a number of degrees of

freedom equal to the number of coordinate functions used. In this report, the containers are assumed cylindrical membranes with flat, rigid bottoms. For simplicity, typical tandem parts of vehicles, consisting of only one or two solid masses and springs and one liquid-filled container, are examined. If one-term approaches are considered, the resulting model can be interpreted as an actual spring-mass model. Because of the rapid decrease of the equivalent modal masses (with increasing mode number), restriction to one-term approaches does not mean considerable loss of accuracy. Comparison of analytical results with experiments provides satisfactory agreement. In this way, the use of large matrices in the final eigenvalue problem can be avoided.

TN D-5425

September 1969

Numerical Integration in a Rigid-Body Trajectory Program. John Edd Moore. Aero-Astroynamics Laboratory. N69-38769.

A comparison is made of the fourth-order Runge-Kutta numerical integration technique with two numerical integration formulas in a mathematical model used for computing rigid body, first-stage trajectories of a Saturn space vehicle. The scope of the problem is to describe the mathematical model and to present a derivation of the differential equations of motion which it comprises; to establish a basis for comparing the integration techniques; and to generate sufficient comparison data to establish which integration technique is the most practical to use in this model from a standpoint of computer run time and accuracy. An analysis of the data showed a fourth-order formula to be the most practical integration technique to use in computer programs of this type from a standpoint of computation time, accuracy, and stability of the solution.

TN D-5426

October 1969

Rotating Space Station Stabilization Criteria for Artificial Gravity. Carl A. Larson. Astronautics Laboratory. N69-39210.

An understanding of man's dependency on a gravity field is an area where research has yielded limited results, thereby, prompting space station designers to include provisions for providing an

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artificial gravity field capability. A roto-graviv environment, though satisfying certain of man's physiological requirements, can introduce other complicated requirements. Therefore, this investigation was devoted to obtaining, for space station designers, insight into how man's physiological tolerances and range of adaptability can be used to define operational and configurational design criteria.

TN D-5439

October 1969

Final Report: Saturn V, S-IVB Panel Flutter Qualification Test. J. J. Nichols. Astro-nautics Laboratory. N69-38808.

This report gives results obtained from an evaluation of data taken during the Saturn S-IVB forward skirt flutter test conducted at Arnold Engineering Development Center, Tullahoma, Tennessee, during the period from October 30 through November 6, 1967. A fullscale segment of the S-IVB stage forward skirt was tested to determine the flutter characteristics of the thin skin panels bounded by internal rings and external stiffeners.

Since the skin panels were subjected to more severe flutter conditions than would be expected during flight, it is concluded that the S-IVB stage panel flutter is not critical for all Saturn IB and Saturn V published flight trajectories.

TN D-5469

November 1969

A Theoretical and Experimental Study of Unsteady Flow Processes in a Ludwig Tube Wind Tunnel. John D. Warmbrod. Aero-Astro-dynamics Laboratory. N70-10710.

The application of a numerical procedure generally referred to as the method of characteristics to calculate the nonsteady discharge of compressed air through a Ludwig tube type of wind tunnel is presented. The mathematical model that underlies the numerical procedure assumes the flow through the wind tunnel to be time-dependent, one-dimensional, and compressible. The theoretical results presented were obtained from a computer program developed for this problem. Diaphragm rupture, variations in cross-sectional areas, contact surfaces, shock waves, and the interaction and reflection of contact surfaces and shock waves are

included in the computer program. The theoretical results are compared with static and total pressures measured in a Ludwig tube wind tunnel that has recently begun preliminary operation at MSFC. Good agreement between theory and experiment is achieved for the cases in which the flow in the test section was subsonic. Only fair agreement between the one-dimensional theory and experiment was achieved in the supersonic test sections. This difference is believed to be attributed to the two-dimensional character of the flow and the extensive flow separation that occurs in the supersonic nozzles during the early development of the flow.

TN D-5470

November 1969

Numerical Solution of Differential Equations Using Obrechhoff Corrector Formulas. Joseph S. Rosen. Computation Laboratory. N70-10685.

The solution of differential equations by predictor-corrector formulas which give high-order accuracy is considered. The corrector formulas are obtained from an extension of Obrechhoff's formula. These formulas are characterized by the fact that they contain higher derivatives of the unknown function $y(x)$, and if their derivatives are readily determinable, fewer terms are needed for a given order as compared to other multistep methods which do not involve derivatives. Corresponding to these corrector formulas, certain new predictor, or extrapolation, formulas are suggested. These formulas also make use of the derivatives used in the corrector formulas.

TN D-5517

December 1969

Flowmeter for Space. T. F. Morris. Quality and Reliability Assurance Laboratory. N70-12393.

This report discusses the development of a mass flowmeter designed for use in outer space. The purpose of the flowmeter is to measure flow-rates from purges and collected leaks at leak ports, on aerospace hardware, discharging into a space environment. One of the most notable features of the flowmeter is the capability to measure the flow of all common gases, such as hydrogen, helium, nitrogen, oxygen, and organic vapors. These gases can be measured over a wide range of flowrates

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(nitrogen, 5 to 2500 sccm) in a space environment of zero-gravity to extremely low pressures, and with temperature extremes of 70° to 500°K.

TN D-5584

February 1970

Longitudinal and Lateral Spectra of Turbulence in the Atmospheric Boundary Layer. George H. Fichtl and George E. McVehil. Aero-Astro dynamics Laboratory. N70-19272.

An engineering spectral model of turbulence is developed with horizontal wind observations obtained at the NASA 150-meter meteorological tower at Cape Kennedy, Florida. Spectra, measured at six levels, are collapsed at each level with Monin coordinates, where the longitudinal or lateral spectral energy density, the surface friction velocity, and the mean wind speed at some height are represented. A vertical collapse of the dimensionless spectra is produced by assuming they are shape invariant in the vertical. An analysis of the logarithmic spectrum in the inertial subrange, at the 18-meter level, implies that the local mechanical and buoyant production rates of turbulent kinetic energy are balanced by the local dissipation and energy flux divergence, respectively.

TN D-5633

February 1970

Shear Layer and Jet Instability in Stratified Media. George H. Fichtl. Aero-Astro dynamics Laboratory. N70-21619.

The stability to small perturbations of shear layer and jet flows in atmospheres with potential temperature is investigated. The problem is reduced to a characteristic value problem for the dimensionless wave frequency which appears in a second-order differential equation with the dependent variable being the horizontal and temporal Fourier transform amplitude of the vertical component of the perturbation momentum vector. Broken-line profiles of jet flow and potential temperature are used in the analysis of this problem.

Integral equations, over the domain of the fluid, which contain both quadratic forms and interfacial contributions, are derived. The interfacial terms vanish for continuous flows, and the theorems of Synge, Howard, and Miles follow. A necessary and sufficient condition for instability is also obtained

for continuous flows; however, its usefulness is compromised by integrands which depend on both the basic state flow and the dependent variable of the governing differential equation.

TN D-5634

February 1970

Discontinuous Flows and Free Streamline Solutions for Axisymmetric Bodies at Zero and Small Angles of Attack. Heinz G. Struck. Aero-Astro dynamics Laboratory. N70-21527.

Except for a few numerical results, nothing has been published, comparable in magnitude to the two-dimensional case, concerning free streamline solutions of the axisymmetric problem. Since the method of using conformal transformations is not available in axisymmetric and three-dimensional flows, it is necessary to adopt approximate methods for the free streamline analysis. In this study an integral equation procedure has been applied to calculate the free streamline flow behind axisymmetric bodies at zero and small angles of attack. The method uses a modified condition for irrotational flow and iterates until the proper streamline location is found. Some of the results obtained by this procedure will be discussed in this report.

TN D-5687

March 1970

Effect of Wall Roughness on the Damping of Liquid Oscillations in Rectangular Tanks. Frank M. Bugg. Aero-Astro dynamics Laboratory. N70-20583.

An experimental investigation was made to determine the effectiveness of wall roughness for damping liquid oscillations in rectangular tanks. Roughness with height equal to 0.427 percent of the wetted wall area gave damping 65 percent greater than the smooth wall damping. Comparisons were made with viscous damping theory.

TN D-5714

March 1970

Stability of Constant-Gain Systems With Vector Feedback. George L. von Pragenau. Astrionics Laboratory. N70-21840.

Vector feedback is defined as a multivariable feedback loop that cannot be opened at a single nodal point. Such systems are treated for the constant-gain

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(linear and time-invariant) case by decomposing such systems into stable subsystems. Ordinary algebraic operations always permit such decompositions. Relatively simple necessary and sufficient conditions result, and stability is analyzed for a closed-loop system in which the input is a disturbance vector and the output is a feedback vector. The proposed method permits a drastic reduction in the dimensions of a problem because of a building block approach. The state-space method is compared and found to be the limit case where the building block formulation assumes its maximum dimensions. Noncontrollable and nonobservable system parts are not nonconsequential because all subsystems are stable. The method presented was successfully used for the stability analysis of the Apollo-Saturn V pogo phenomenon where eigenvalues numbered well over a hundred. This problem was reduced to a single Nyquist plot analysis of a 4 by 4 determinant, demonstrating the power in reducing dimensions of a problem that would have been a formidable task for the state-space approach. The proposed method is generally applicable.

TN D-5722

April 1970

A Study of Radiation Heat Transfer From a Cylindrical Fin With Base Surface Interaction. Charles A. Cothran. Astronautics Laboratory. N70-24257.

This paper presents a study of the temperature distribution of a cylindrical fin transferring heat simultaneously by internal conduction and by surface radiation. The temperature distribution was studied considering the fin only and the more complex case of surface interactions with a circular base. Equations are derived for the temperature distribution of the fin only, the fin with base surface interactions when both surfaces are black, and also the case of gray diffuse surfaces. Solutions to the equations are obtained by numerical techniques, the analysis is verified experimentally, and the results are given.

TN D-5782

May 1970

A Finite Element Approach to the Structural Instability of Beam Columns, Frames, and Arches. Jerrell M. Thomas. Astronautics Laboratory.

A nonlinear stiffness matrix for a beam column element subjected to nodal forces and to a uniformly distributed load is developed from the principle of virtual displacements and the bifurcation theory of elastic stability. Three cases of applied load behavior are considered. The buckling load of a uniform circular arch is calculated as an example.

TN D-5808

June 1970

Nonlinear Fluid Oscillations in a Partially Filled Axisymmetric Container of General Shape. John Ray Admire. Astronautics Laboratory.

A solution for finite-amplitude free oscillations of fluid in a partially filled axisymmetric container is presented. The formulation of the problem resulted in a nonlinear boundary value problem where the nonlinearity occurred in the boundary condition at the free surface of the fluid. The boundary condition at the container wall and the differential equation are linear. The solution is obtained by first linearizing the free surface boundary condition and solving the resulting linear boundary value problem. Then, using the linear solutions, which satisfy the differential equation and the boundary condition at the container wall, a solution is found that satisfies the nonlinear boundary condition at the free surface asymptotically. A solution through the third-order term is developed. Numerical results are obtained for finite-amplitude standing waves near the first antisymmetric linear mode in three axisymmetric vessels. Results are found for three containers, a cylindrical tank with an ellipsoidal bulkhead, a cylindrical tank with a conical bulkhead, and a cylindrical tank with a truncated conical bulkhead.

TN D-5825

May 1970

Method for Calculating Rocket Engine Structural Loads. Jerrell M. Thomas. Astronautics Laboratory.

A method for calculating structural loads in a rocket engine-actuator-vehicle system is developed. The engine is attached to the vehicle by a universal-type gimbal joint and by two actuators. The engine is assumed to be a rigid body. Either the actuator loads or the engine angular acceleration is assumed to be known with a calculation of the other required. A set of algebraic equations is developed from which

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the unknown actuator loads or angular acceleration and the forces and moments at the gimbal point can be calculated.

TN D-5841

June 1970

Leak Detector for Use in Space Environment.
K. W. Woodis. Quality and Reliability
Assurance Laboratory.

The design and manufacturing techniques and tests required to build a leak detector for aerospace hardware are presented. The basic configurations and power requirements are described in detail and the application of a leak detector to aerospace hardware is explained. The report shows that it is electronically, mechanically, and economically feasible to build leak detectors to support orbital inspection and checkout operations of large space stations and interplanetary vehicles.

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TR R-315

July 1969

Low-Order Classical Runge-Kutta Formulas With Stepsize Control and Their Application to Some Heat Transfer Problems. Erwin Fehlberg. Computation Laboratory. N69-30753.

New first-, second-, third-, and fourth-order Runge-Kutta formulas with stepsize control procedures are derived. These new Runge-Kutta formulas have very small truncation errors; therefore, their application reduces computer time substantially. These low-order formulas are also applicable to parabolic partial differential equations as encountered in heat transfer problems. Examples for ordinary and partial differential equations are presented. In these examples the new formulas require only a fraction of the computer time of conventional formulas.

TR R-321

September 1969

Calibration of Pegasus and Explorer XXIII Detector Panels. Robert J. Naumann, David W. Jex, and Clyde L. Johnson. Space Sciences Laboratory. N69-36985.

Several series of tests were conducted to establish the ballistic limit for the various Pegasus and Explorer XXIII detector panels. In these tests, extremely small, carefully sized beads were launched in a velocity range that included the perforation threshold. By varying the size, a ballistic limit curve was established on a mass-velocity plot. Such curves were then extrapolated to average meteor velocity to obtain the characteristic mass for each sensor. Additional tests were conducted on the Pegasus panels using the Pegasus test tool, which contains the identical electronic circuitry used by the flight models. These tests included hot and cold tests, thermal cycling after impact, oblique impacts, etc. Several types of anomalous events noted in the operation of the spacecraft could be simulated in this manner. Some indication of the dependence of detection threshold upon size was obtained, and the implications of these tests in the analysis of the results will be discussed.

TR R-322

October 1969

Determination of Meteoroid Environments from Photographic Meteor Data. Charles C. Dalton. Aero-Astroynamics Laboratory. N69-39336.

A mathematical model is used to represent Öpik's 1958 physical theory of meteors in a form convenient for programming the computation of meteoroid photometric mass values. Subsamples of 333 photographic meteors from McCrosky and Posen's sample are selected with respect to magnitude scaled for minimum velocity. A statistical comparison between the 1958 Öpik result and the 1933 Öpik provisional result, the Harvard-Meteor project basis for mass values, is given. Meteor and orbit parameter distributions and mean cumulative flux in absolute units for mass, momentum and energy are given separately for the terrestrial influx and for the lunar and interplanetary vehicle onfluxes.

TR R-335

April 1970

The One-Dimensional Time-Dependent Interaction of a Satellite with the Ionospheric Plasma. Nobie H. Stone and John W. Sheldon. Space Sciences Laboratory. N70-24381.

A theoretical study and numerical solution are presented for the one-dimensional, time-dependent problem of a streaming, nonuniform, fully ionized, collisionless plasma impinging upon a stationary, flat conducting plate oriented perpendicularly to the flow of the plasma. The problem is intended to model the interaction of the frontal surface of a satellite with the ionospheric plasma. The time-dependent equations that describe the one-dimensional problem are established, and from them the steady-state equations are derived. These steady-state equations are solved for the spatial dependence of the electric field, which is seen to be a reasonable form and to agree with previous work on more complicated configurations. Finally, a numerical method of solving the time-dependent equations is presented and applied to two special cases: (1) for a plate that has no initial charge, and (2) for a system that is initially in an equilibrium state and proceeds to a second equilibrium state because of a step transition in the plasma parameters.

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(Abstracts for these reports may be obtained from STAR.)

CR-1410	August 1969	CR-61296	September 1969
Investigation of the Turbulent Wind Field Below 150-Meter Altitude at the Eastern Test Range. NAS8-21140. Pennsylvania State University. N69-35570.		STARLAB: An Orbiting Space Technology Applications and Research Laboratory. Final Report. NSR 01-003-025. Auburn University Engineering Systems Design Summer Faculty Fellows.	
CR-1418	September 1969	CR-61297	September 10, 1969
Ground Wind Characteristics at Kennedy Space Center. NAS8-21178. Cornell Aeronautical Laboratory, Inc. N69-36222.		An Application of the Multivariate Extended Poisson Distribution in 2 Times 2 Contingency Tables. NAS8-11175. University of Georgia. N69-36015.	
CR-1465	January 1970	CR-61298	August 1969
Numerical Calculations of Viscous Compressible Fluid Flow Around a Stationary Cylinder. NAS8-18034. Applied Theory, Inc. N70-17371.		A Preliminary Upper Atmospheric Density Model Derived From Satellite Drag Density Data. NAS8-30513. Lockheed Missiles and Space Co. N69-37255.	
CR-1466	February 1970	CR-61299	September 1969
Numerical Calculations of Viscous Compressible Fluid Flow Over a Flat Plate and Step Geometry. NAS8-11400. Applied Theory, Inc. N70-18252.		Numerical Analysis of Shells, Volume I: Unsymmetric Analysis of Orthotropic Reinforced Shells of Revolution. NAS8-21113. Grumman Aircraft Engineering Corp. N70-12498.	
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Numerical Calculation of Viscous Compressible Fluid Flow Around an Oscillating Rigid Cylinder. NAS8-18034. Applied Theory, Inc. N69-38782.		Numerical Analysis of Shells, Volume II: User's Manual for STARS-II — Shell Theory Automated for Rotational Structures-II-Digital Computer Program. NAS8-21113. Grumman Aircraft Engineering Corp. N70-12499.	
CR-1529	May 1970	CR-61301	September 1969
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Trajectories for the 1976-1980 Grand Tour Opportunities, Volume I: Graphic and Summary Trajectory Data. NAS8-20082. Lockheed Missiles and Space Co. N70-15089.		AS-507 H Mission Launch Vehicle Operational Trajectory for a December 1969 Launch. NAS8-20082. Northrop Corp.	
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Peak Wind Statistics Associated With Thunderstorms at Cape Kennedy, Florida. NAS8-20082. Lockheed Missiles and Space Co. N69-37958.		Static and Dynamic Calibrations of MSFC's Plug-Nozzle Special Test Section (Redesigned 1967). NAS8-20082. Nortronics. N70-11662.	
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Trajectories for the 1976-1980 Grand Tour Opportunities, Volume II, Trajectory Data. NAS8-20082. Lockheed Missiles and Space Co.		Evaluation of Odyssey-1 Orbital Aerodynamic Experiment Package. NAS8-20082. Northrop Corp. N70-15057.	
CR-61306	August 1969	CR-61314	September 1969
Prediction of Geomagnetic Activity. NAS8-30156. McDonnell-Douglas Astronautics Co. N69-39662.		Computer Program for Crossed-Beam Studies of Clear Air Turbulence, Program Description (MLTCOR). NAS8-21300. IIT Research Inst. N70-12025.	
CR-61307	September 1969	CR-61315	September 1969
Experimental Material Handling Device. Final Report. NAS8-30069. Martin-Marietta Corp.		Bolt Installation and Removal Tool. NAS8-30026. Aerojet-General Corp.	
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Cape Kennedy Peak Wind Profile Probabilities for Levels From 10 to 150 Meters. NAS8-20082. Lockheed Missiles and Space Co. N70-12204.		Long Range Solar Flare Prediction. NAS8-21436. Denver Research Inst. N70-14569.	
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Laboratory Development of a Satellite Experiment for Gas-Surface Interaction Studies. NAS8-21450. University of Michigan.		Solid State Radiographic Image Amplifiers. Part B. NAS8-21206. Electronic Tube Div. of Westinghouse Electric Corp.	
CR-61321	November 1969	CR-61329	June 1970
User's Manual for RAVFAC, a Radiation View Factor Digital Computer Program. NAS8-30154. Lockheed Missiles and Space Co.		Space Radiation Hazards To Project Skylab Photographic Film. NAS8-24514. Lockheed-Georgia Co.	
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